

DESIGN AND MANUFACTURING OF TEXTILE STRUCTURAL COMPOSITES
(TXL766)

MINOR I (20%)

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Date: 5 Feb 2020

Venue: LI1318

Time: 1 h

1. Select examples (from box) for the following: (4)

- Isotropic body
- Homogeneous body
- Homogeneous body that is not isotropic
- Isotropic body that is not homogeneous

- 1. Steel rod
- 2. Steel rod heated from one end
- 3. Wood
- 4. Wood with knots

2. From the values provided in Table 1, answer the following:

- Identify the strongest and the weakest fibre for the composite application. What is the ratio of their strength? (2+2)
- Identify the fibre with the minimum or the maximum stiffness (ability to resist deformation) for the composite application. What is the ratio of their stiffness? (2+2)
- Select the fibre and the matrix combination for a unidirectional continuous composite to achieve the maximum specific longitudinal modulus at $(V.F.)_{\text{fibre}} = 50\%$. Find its value. (2+2)
- A unidirectional continuous hybrid composite is made from the epoxy matrix (volume fraction 60%) which is reinforced with Kevlar (20%) and Boron (20%) fibres. Find the strength of the composite in its longitudinal direction. (Assume: A composite fails as soon as any fibre breaks). (4)

Bonus (+5)

A unidirectional continuous composite of Carbon-Epoxy (Fibre Volume Fraction = 50%) is under stress condition as shown. Find the strains in longitudinal and transverse direction.

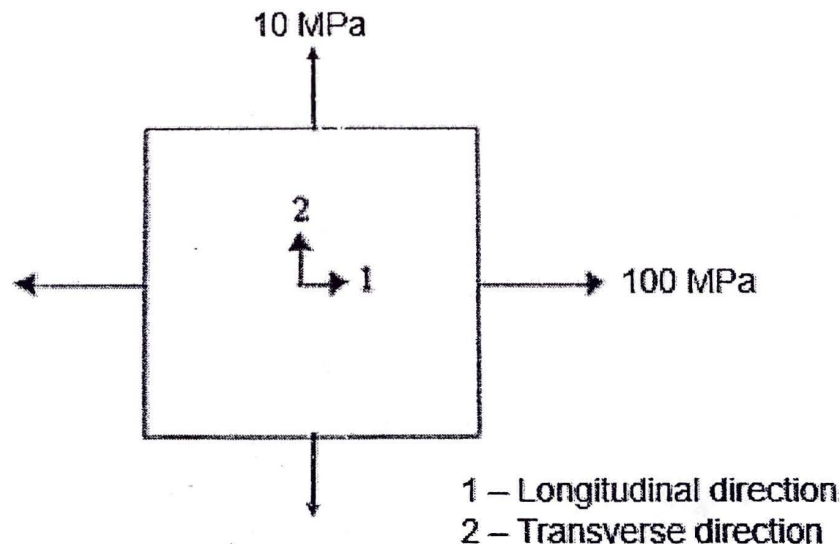


Table 1: Mechanical Properties of Textile Fibres and Matrix for Composite

Materials	Density (Kg/m ³)	Modulus (GPa)	Breaking Stress (GPa)	Breaking Strain (%)	Poisson's Ratio
<u>Carbon</u>	2100	500	7	3	0.3
<u>Glass</u>	2600	40	1	5	0.25
<u>Kevlar</u>	1440	300	3.6	0.12 7	0.35
<u>Boron</u>	2380	200	1.4	0.07 4	0.2
<u>Steel</u>	7800	207	.99	2	0.2
<i>Polyester</i>	1500	4	0.15	2	0.3
<i>Epoxy</i>	1400	6	0.1	0.16 6	0.34
<i>Polypropylene</i>	920	2	0.0414	15	0.36
<i>Polystyrene</i>	1060	5	0.069	2.5	0.34
<i>Nylon 6</i>	1140	3	0.079	20	0.37

Assume Fibre and Matrix as a Isotropic Material

Relationship of σ/ϵ for a Thin Lamina (Orthotropic) under Plane Stress Condition

$$\begin{bmatrix} \epsilon_1 \\ \epsilon_2 \\ \gamma_{12} \end{bmatrix} = \begin{bmatrix} \frac{1}{E_1} & -\frac{\nu_{12}}{E_1} & 0 \\ -\frac{\nu_{21}}{E_2} & \frac{1}{E_2} & 0 \\ 0 & 0 & \frac{1}{G_{12}} \end{bmatrix} \begin{bmatrix} \sigma_1 \\ \sigma_2 \\ \tau_{12} \end{bmatrix}$$